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# **Feasibility of Third World Advanced Ballistic & Cruise Missile Threat Volume 1: Long Range Ballistic Missile Threat**

**Presented to**  
**67th MORS Symposium**  
**Working Group 4: Air and Missile Defense, TMD Threat**

**June 24, 1999**

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# REPORT DOCUMENTATION PAGE

*Form Approved  
OMB No. 0704-0188*

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1. REPORT DATE (DD-MM-YYYY) 24-06-1999		2. REPORT TYPE Technical Study		3. DATES COVERED (From - To) 1998-1999
4. TITLE AND SUBTITLE  Feasibility of Third World Advanced Ballistic and Cruise Missile Threat: Volume 1: Long Range Ballistic Missile Threat				5a. CONTRACT NUMBER
				5b. GRANT NUMBER
				5c. PROGRAM ELEMENT NUMBER
				5d. PROJECT NUMBER
				5e. TASK NUMBER
				5f. WORK UNIT NUMBER
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES)  National Defense Industrial Association (NDIA) 2111 Wilson Blvd., Suite 400 Arlington, VA 22201				8. PERFORMING ORGANIZATION REPORT NUMBER
9. SPONSORING / MONITORING AGENCY NAME(S) AND ADDRESS(ES)  Naval Sea Systems Command 2531 Jefferson Davis Highway NC#2 Arlington, VA 22242				10. SPONSOR/MONITOR'S ACRONYM(S) OPNAV PEO (TSC)
				11. SPONSOR/MONITOR'S REPORT NUMBER(S) N/A
12. DISTRIBUTION / AVAILABILITY STATEMENT  The NDIA Study "Long range Ballistic Missile Threat" has been reviewed and is approved for public release; distribution is unlimited.				
13. SUPPLEMENTARY NOTES  Abbreviated study materials presented at 67 <sup>th</sup> Military Operations Research Society Symposium at US Military Academy 24 June 1999.				
14. ABSTRACT  "Feasibility of Third World Long Range Ballistic Missile Threat" presents the results of a United States aerospace industry study which assesses the likelihood of a Third World country developing a long range (3,000-10,000 km) ballistic missile (LRBM) system and the estimated time to field it. A 10,000-km range ballistic missile launched from North Korea, for example, can reach the western and central regions of the United States. Likewise, a 5,000-km range missile launched from Iran could reach cities throughout Western Europe. The study contains four technical sections: historical developments and technology migration, trends in Third World ballistic missile weaponry, threat development on a compressed schedule, and candidate LRBMs configurations. The report is unclassified because it drew exclusively upon unclassified sources of information. The study traces the history of the LRBMs threat from Germany's V-2 rockets of World War II to the present. It shows how Third World countries could quickly field and launch LRBMs with technical assistance and components imported from developed nations. The study examines five different options by which a Third World country could achieve a long range ballistic missile capability: Buy a long range ballistic missile; Buy and convert an available space launch vehicle; Cluster or stack existing tactical missiles as boosters; Design and build a booster and use existing tactical missile for upper stage; or Design and build an entire missile. The flight stability and performance characteristics of feasible Third World missile configurations are based on industry experience in the design of missile systems and verified by the use of standard engineering analysis tools and missile flight simulations. The report contains the estimated time required for a Third World nation to develop each option as measured from program start to first launch. The study concludes that the time between detectable evidence of a country's missile development program until the missile is fielded may be less than the time needed to build defenses against it.				
15. SUBJECT TERMS  Ballistic missiles, history, technologies, guidance, warheads, propulsion, employment, threats, advanced weapons, proliferation, performance simulation, design synthesis				
16. SECURITY CLASSIFICATION OF: UNCLASSIFIED		17. LIMITATION OF ABSTRACT Unlimited	18. NUMBER OF PAGES 48	19a. NAME OF RESPONSIBLE PERSON Robert E. Woodside
a. REPORT UNCLASSIFIED	b. ABSTRACT UNCLASSIFIED	c. THIS PAGE UNCLASSIFIED		19b. TELEPHONE NUMBER (include area code) (253) 773-4164



**Feasibility of Third World  
Advanced Ballistic & Cruise Missile Threat  
Volume 1: Long Range Ballistic Missile Threat**

Prepared By  
**Systems Assessment Group**  
**NDIA Strike, Land Attack and Air Defense Committee**

Released  
**November 1998**

# **Study Approach**

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- **Review lessons learned from historical missile development programs**
  - WW II Germany, North Korea, Iraq
  - Development time and motivation
  - Strategic objectives for LRBMs with WMD warheads
- **Assess potential range growth of evolving family of Third World TBM threats to include LRBMs with WMD warheads (3,000km - 10,000 km range)**
- **Conduct Third World Threat Analysis & Development Timelines**
  - Technology assessment
  - Development forecast
- **Assess Candidate LRBMs Configuration Alternatives**
  - Space launch vehicle conversion
  - TBM stacking / clustering

# **Outline**

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- **German Ww II Missile Development and Technology Migration**
- **Trends in Third World Ballistic Missile Weaponry**
- **Threat Development on a Compressed Schedule**
- **Candidate LRBM Configurations**
- **Summary**

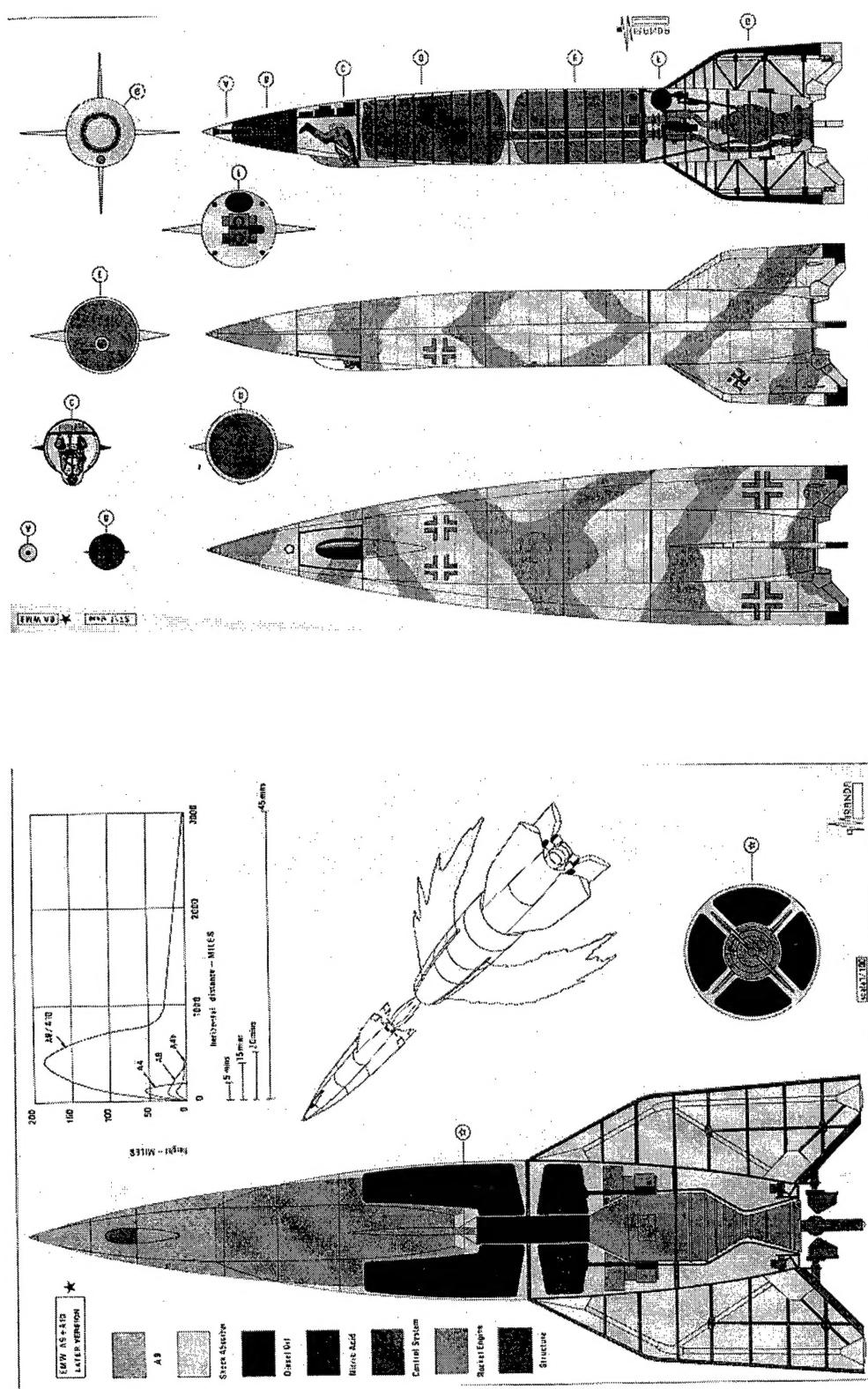
# Key Ingredients of the German Program

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- Political/Cultural Climate
  - Rocket development not constrained by Versailles treaty
  - Autocratic rule
  - Warring factions within the Nazi party for control of ballistic missile program
  - 1920's Weimar Republic had strong interest in rocketry and space flight
  - Use of slave labor
- Technological and Strategic Surprise
  - Early recognition of the value of ballistic missiles for maximizing surprise (1929)
  - Extreme secrecy of the program - first large "black" program
  - Suppression of German amateur rocket societies for security reasons (1933-34)
- Speed of Development
  - National priority / significant funding resources available
  - Excellent domestic industrial skill base to develop required technologies
  - Rapid prototyping using technology demonstrators (Test / Fail / Fix / Re-Test...)

*Many of these key ingredients are found in Third World programs today*

# German Intercontinental Missile Design



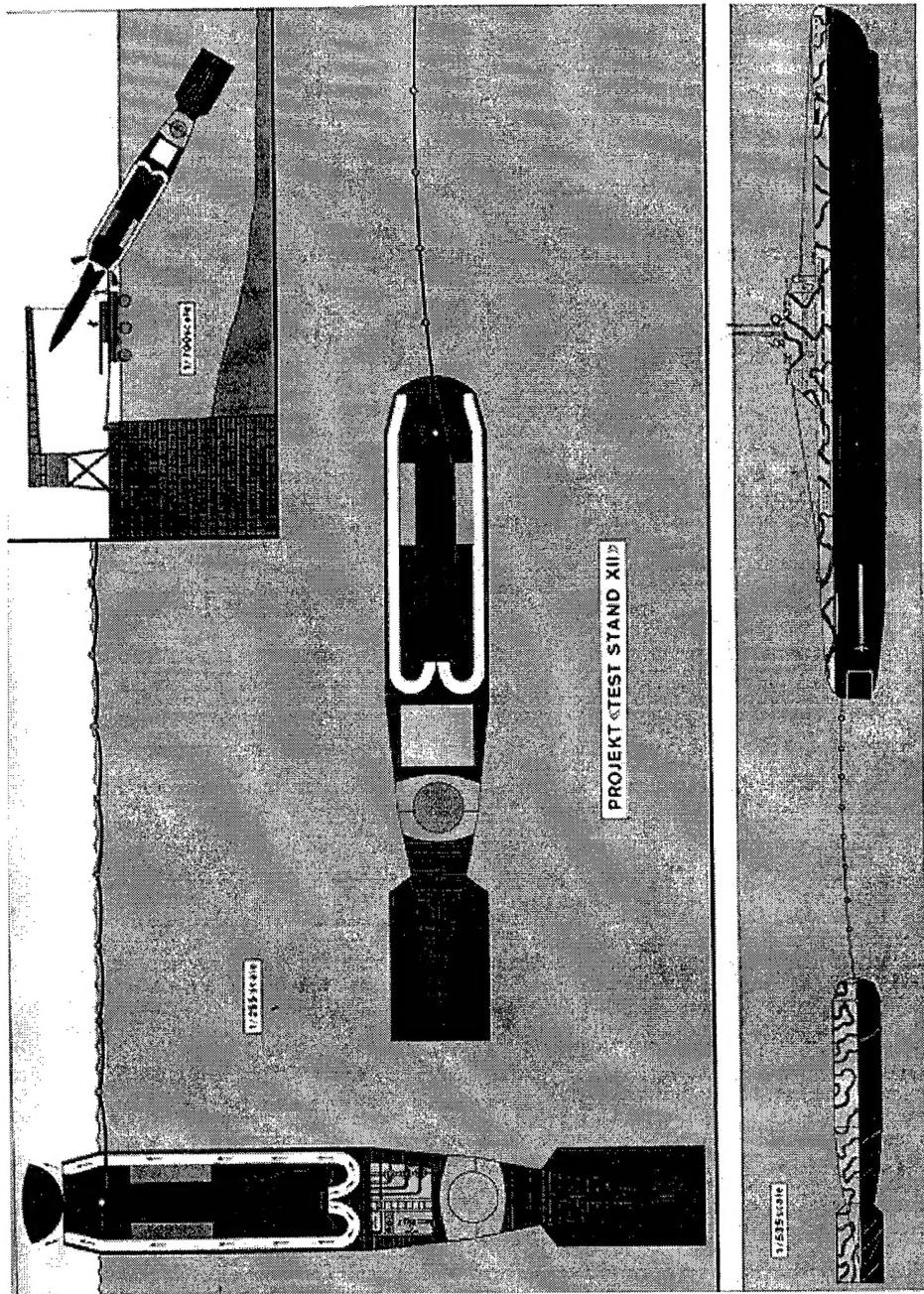
**Manned A9**

Source: "Secret Wonder Weapons of the Third Reich", J. Miranda, P. Mercado, Schiffer Military/Aviation History, 1996

**A9/A10**

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# Alternative Concept for Attacking North America

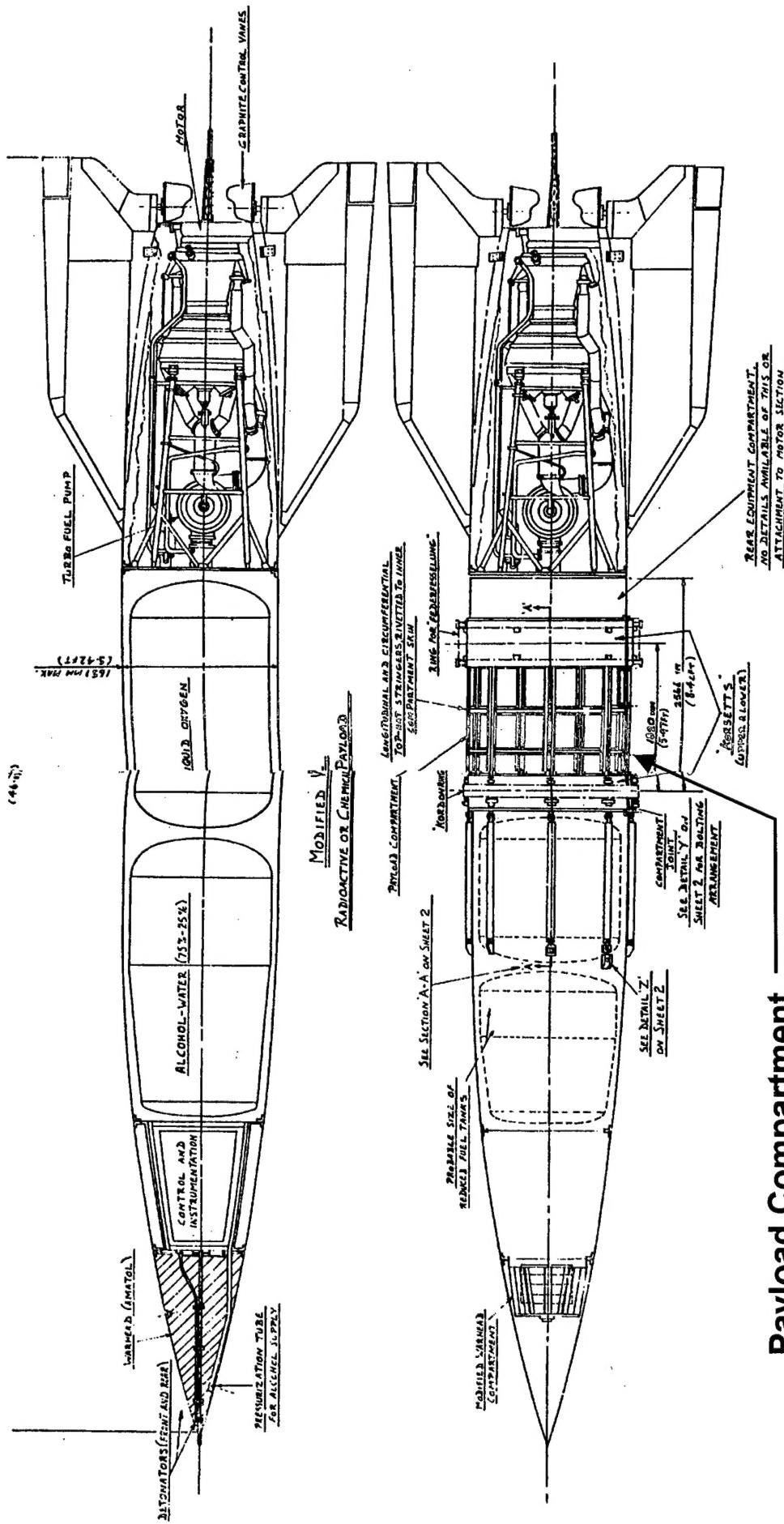


## Project Lifesest

Source: "Secret Wonder Weapons of the Third Reich", J. Miranda, P. Mercado, Schiffer Military/Aviation History, 1996

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# Design for Use of Chemical/Radiological Payloads

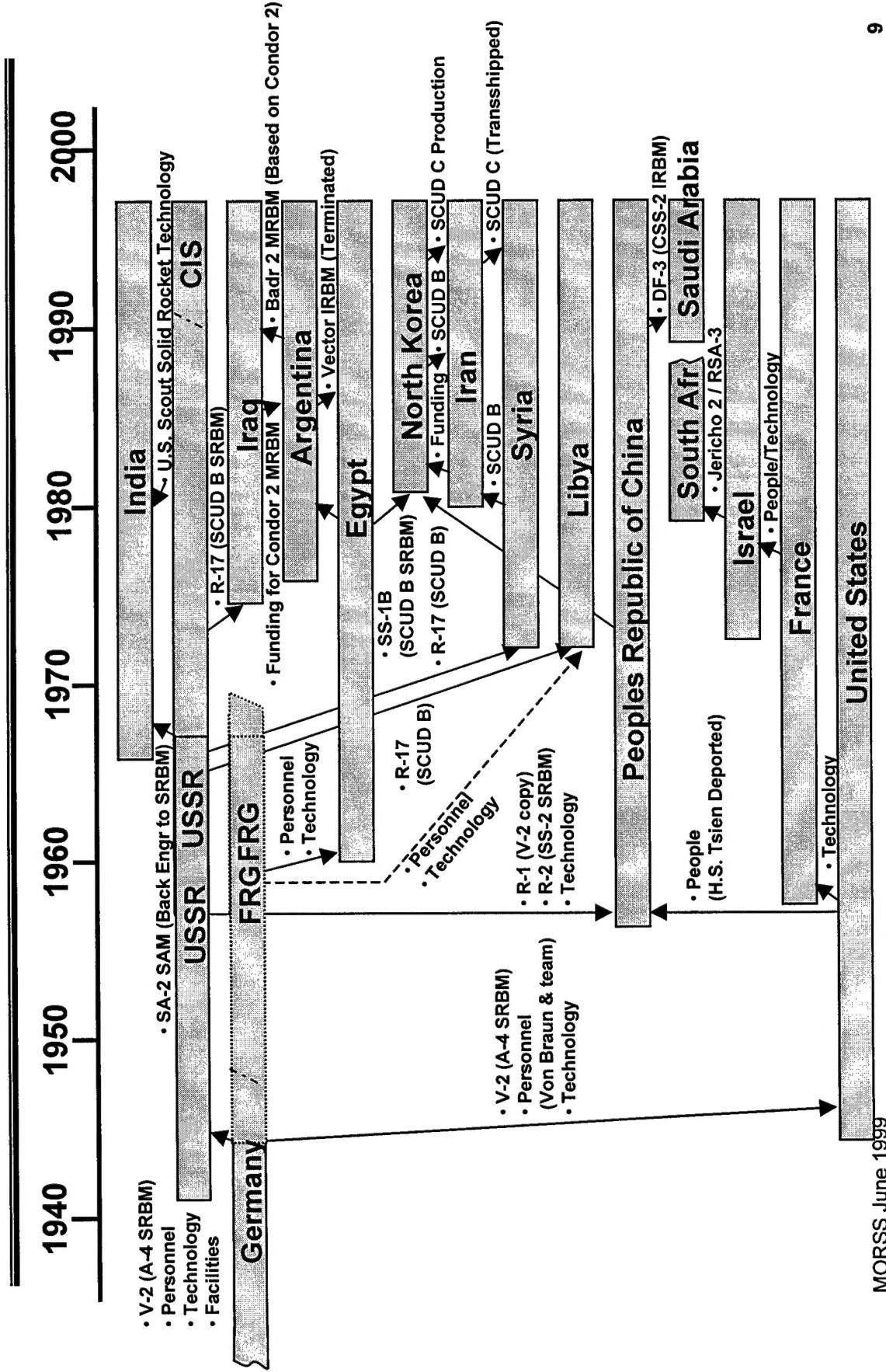


## Payload Compartment

Source: "Vengeance, Hitler's Nuclear Weapon. Fact or Fiction?", Phillip Henshall, Alan Sutton Publishing Ltd., 1995

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# Genealogy of Technology Transfer After WW II



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# International Technology Transfer

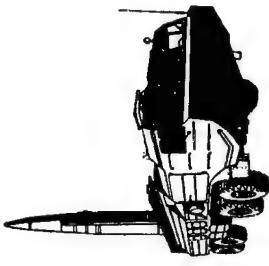
## Technology Suppliers

	Israel	India	Egypt	Iraq	Pakistan	Iran	Libya	Saudi Arabia	Syria
Trained Personnel/Advisors	NL, US	AT, DE, ES, CH, SU, US	AR, AT, BE, CL, EG, FR, DE, LI, KP, CN, CH, US, SU	CN	KP	DE	DE, PK, CN, US		
Reentry Vehicles		DE, CH, US SU	CL, EG, FR, DE, LY, US	CN	CL				
Ballistic missile Propulsion			AT, BE, FR, DE, IT, LI, CH, US		CN, KP			CS, IR, IT LY, SU	
Guidance/Navigation	ZA, TW US		BR, FR, DE, US		HK, JP			IL, CN	
Flight Controls			BR, DE		HK, JP				
Production Assistance	FR, DE NL, US	AR, DE, KP SA, CH, US	AR, AT, BE, BR, EG, FR, DE, IT, MC, KP CN, CH, GB, US, SU	DE, CN	KP, CN	DE	DE	CS, DE, LY KP, SA, SU	
Materials Manufacture		US	DE, LI, GB, US		CA, FR, CH, US				
Computers	US	SA, CH, US	DE		NO, GB			US	
Testing/Ranges	ZA			MR					

Notes: (1) This information was extracted from the International ballistic missile Proliferation Project database, compiled by the Monterey Institute of International studies under the direction of Dr. William C. Potter and Dr. Edward J. Laurance. It is a compilation of open source material and covers only reports of actual deliveries and transfers since 1989. The table does not include information relating to proposals, offers, negotiations, or orders (unless clear transfers have resulted).

(2) The two-letter codes used in this table are the American National Standard Institute (ANSI) international country codes, defined as follows: AR - Argentina, AT - Australia, BE - Belgium, BR - Brazil, CA - Canada, CH - Switzerland, CL - Chile, CN - China, CS - Czechoslovakia, DE - Germany, EG - Egypt, ES - Spain, FR - France, GB - United Kingdom, HK - Hong Kong, IL - Israel, IR - Iran, IT - Italy, JP - Japan, KP - North Korea, LI - Liechtenstein, LY - Libya, MC - Monaco, MR - Mauritania, NL - Netherlands, NO - Norway, PK - Pakistan, SA - Saudi Arabia, SU - Soviet Union, TW - Taiwan, US - United States, ZA - South Africa.

# Iraqi Ballistic Missile Programs



- 1970s -- Iraq receives first Scud-B ballistic missiles from USSR (813 imported by 1990)
- 1982 -- First Iraqi Scud-B attack on Iran
- 3 August 1987 -- Al Husayn tested (500 km range)
- February - April 1988 -- 189 Al Husayn missiles fired in “war of the cities”
- 25 April 1988 -- Al Abbas missile fired on Tehran (860 km)
- 1980-1988 -- Total of 361 Scud-B and Al Husayn missiles fired in Iran-Iraq war
- 5 December 1989 -- Successful launch of Al Abid SLV
- 7 December 1989 -- Tammuz 1 SLV announced (2,000 km range)
- 1991-- Desert Storm saw 88 Al Husayns fired
- 1997 -- Iraq announces Al Hamid ballistic missile (150 km range)

## **Historical Parallel -- 1930s vs. 1990s**

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- **Missile Technology Control Regime no more effective than Versailles Treaty at limiting technology spread**
- **MTCR does not “control” anything. Its signers agree to exercise “restraint”.**
- **Ineffective at stopping:**
  - Transfer of missiles and components from non-signatory states (e.g., North Korea and China)
  - Emigration of scientists, engineers and technicians
- **Technical education in sciences and engineering which provides missile skills**

# **Outline**

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- Trends in Third World Ballistic Missile Weaponry
- Threat Development on a Compressed Schedule
- Candidate LRBMs Configurations
- Summary

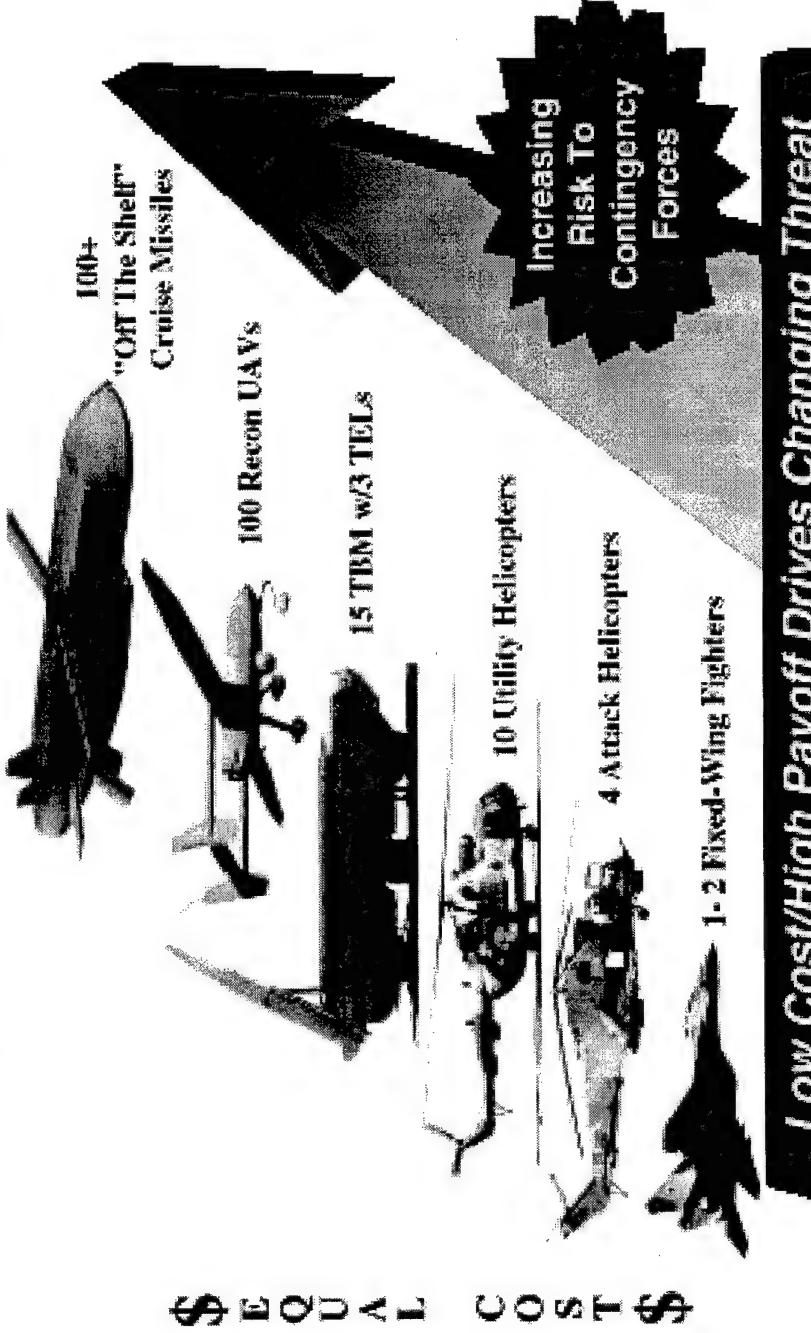
## Third World View of Effective Responses to U.S. Technical Superiority

- Cannot Defeat U.S. / Allies on Battlefield: Land, Sea, Air
  - Persian Gulf War showed failure of direct confrontation
- Development of Hi-Tech Conventional Forces Constrained
  - Affordability
  - Technological Availability
  - National Leadership and Power Structures
- Aggressor Nations Must Develop Asymmetric Strategies & Forces to Counter U.S./Allies

Mission	Objective
- Threaten/Attack Population Centers	- Prevent Creation and Maintenance of Coalitions
- Threaten/Attack U.S. Forward-Deployed Forces and Allied Bases	- Deny Facilities/LOC's to U.S./Allies
- Threaten/Attack Air and Sea Ports of Entry	- Prevent Entry of U.S./Allied Forces
- Threaten/Attack Naval and Amphibious Operations	- Raise Risk to Unacceptable Levels

## MISSILES -- Cost-Effective Force Decision

GIVEN \$50M, ANY ADVERSARY COULD BUY...



*Low Cost/High Payoff Drives Changing Threat*

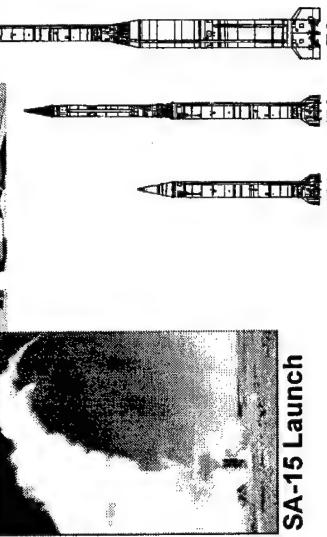
Source: US Army ADA School, Air and Missile Defense Master Plan (threat section), September 1996

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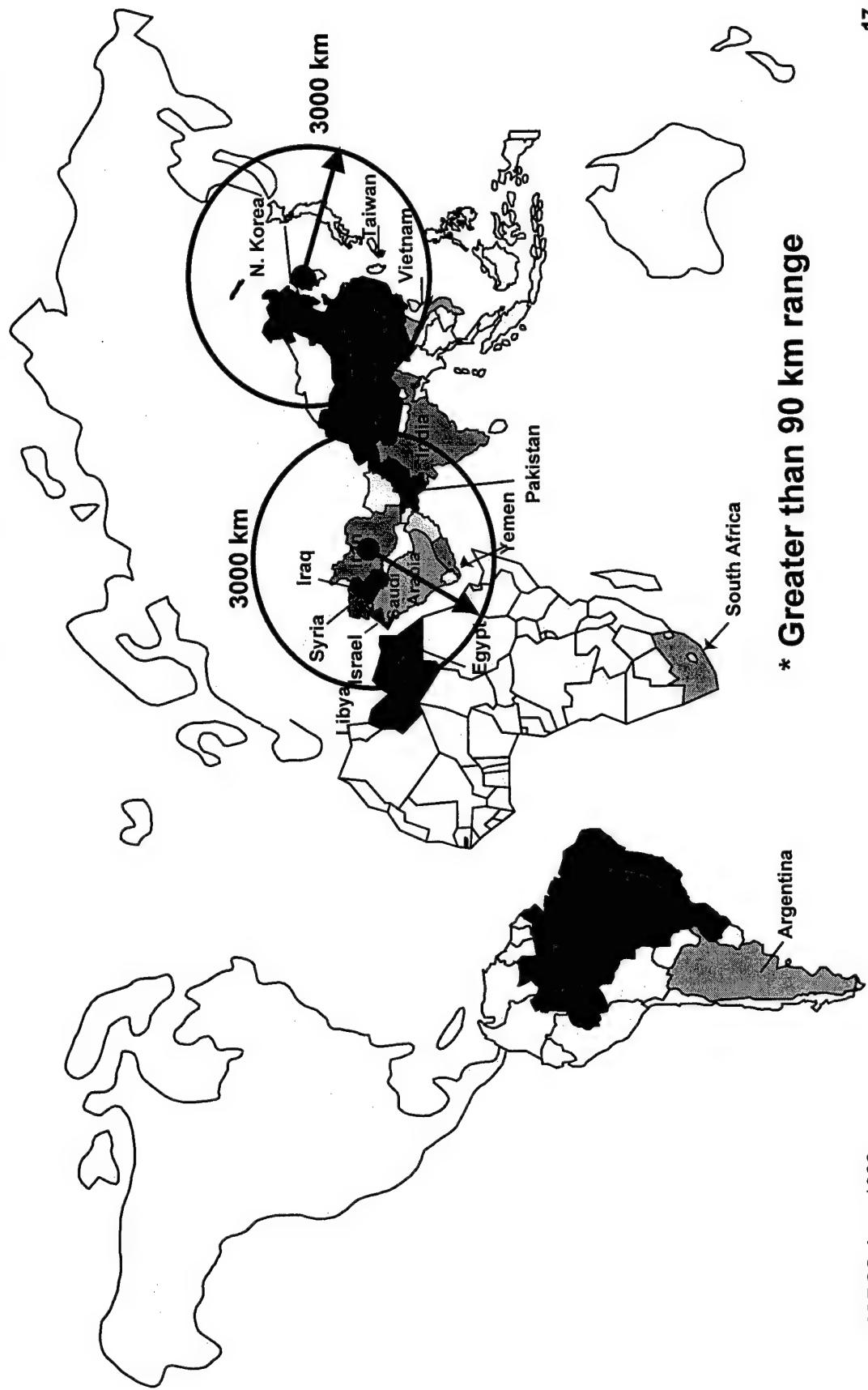
# MISSILES -- Third World Weapons of Choice

- Robust Force Structure
  - Cruise and Ballistic missiles
  - Integrated Air Defense Systems
  - Aircraft with long-range missiles
- Missiles -- Most Cost Effective Investment
  - High speed improves penetration of defenses
  - Onboard seekers & GPS guidance for accuracy
  - WMD delivery meets military/political goals
  - Existing technology conceals development
- Improved Targeting
  - Increased access to space surveillance
  - Modern mobile communications coordinate dispersed launches
  - Computer flight planning allows pre-programmed attack profiles
  - GPS navigation updates enables cheap precision

Su-34 & Missile Armaments



## Third World Countries with Ballistic Missiles \*



# Third World Indigenous Capabilities

	Israel	India	Egypt	Iraq	Pakistan	Iran	Libya	Saudi Arabia	Syria
Human Resources	●	●	●	●	●	●	○	○	○
Ballistic missiles	●	●	●	●	●	●			
Space	●			○	●				
Artillery	●	●	●	●	●	●	○	○	●
Ordnance	●	●	●	●	●	●			
Aircraft	●	●	●	●	●	●			
Electronics	●	●	●	●	●	●			
Test Resources	●	●	●	●	●	●	○	○	

- Substantial capability:
  - Large numbers of experts and technicians
  - Indigenous manufacturing
  - Instrumented test ranges
- Modest capability:
  - Some experts and technicians
  - Spare parts manufacturing
  - Component test/ground test
- Rudimentary capability:
  - Few experts or technicians
  - Repair and refurbishment
  - Limited test facilities

Reference: "Rest of World (ROW) Response to GPALS:  
Technology and Industry Bases", Strategic Defense Initiative Organization, June 1992

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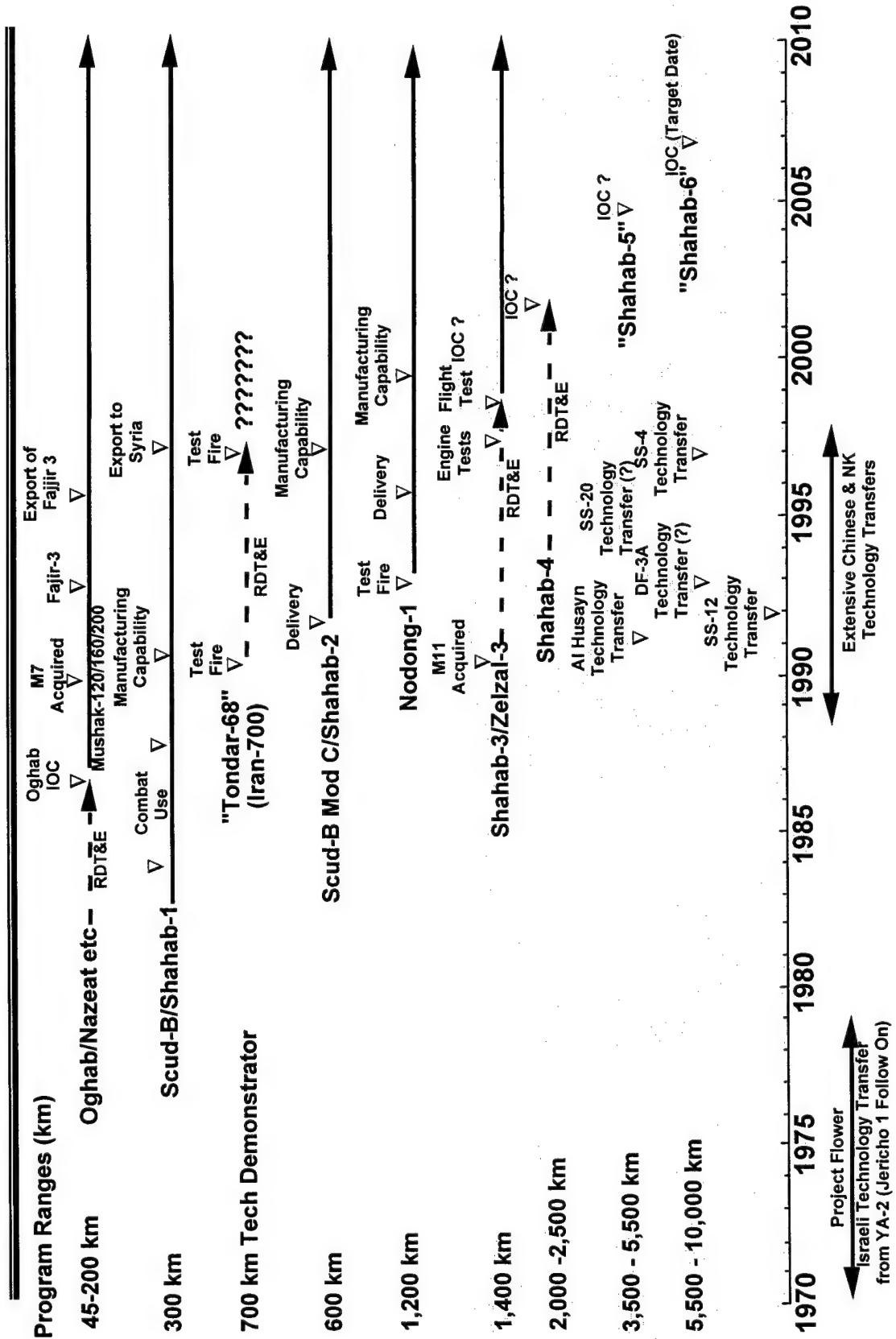
## Current Third World Ballistic Missile Capabilities

Country	Missile System	Status	Propulsion	Launch Weight (kg)	Payload Weight (kg)	Range (km)
China	CSS-1 (DF-2A)	1966 IOC	L	26,000	1500	1250
	CSS-2 (DF-3)	1971 IOC	L/L	64,000	2000	2650
	CSS-2 (DF-3A)	1986 IOC	L/L	64,000	2150	2800
	CSS-3 (DF-4)	1978 IOC	L/L	82,000	2200	4750
	CSS-4 (DF-5A)	1986 IOC	L/L	183,000	3200	13000
	CSS-5 (DF-21A)	1987 IOC	L/L	14,700	600	1800
	DF-25	Reported Stopped	S/S/S		2000	1700
	DF-31	1996 IOC	S/S/S		700	8000
	DF-41	Under Dev.	S/S/S		800	12000
	Agni I	Flight Tested	S/L	16,000	1000	>2000
India	Agni II	Dev. Initiated	S/S		1000	3000
	Shihab 3	Flight Tested	L		750	1700
	Shihab 4	Early Stages of Dev.	S		1000	3000
	Shihab 5	Engaged Effort				3000+
	Labour 1 (No Dong 1 import)		L			
Iran	Labour 2 (No Dong 2 joint effort)	In Dev.	L/L			

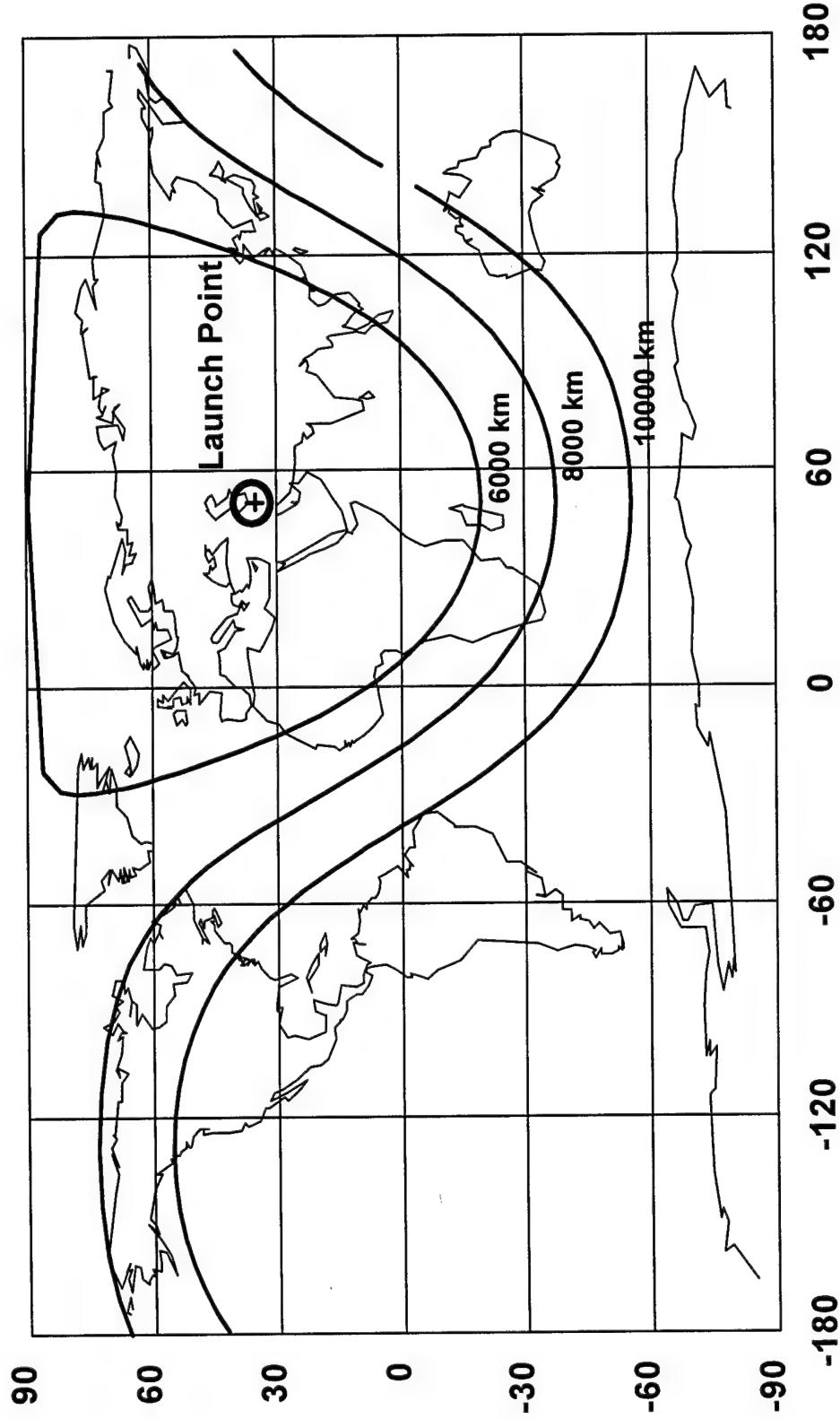
## Current Third World Ballistic Missile Capabilities (Cont)

Country	Missile System	Status	Propulsion	Launch Weight (kg)	Payload Weight (kg)	Range (km)
Iran (cont.)	Tondar 68 (Iran 700)	Dev. Prototype	L or S	500	500	700-1000
	Zelzal -3 (modified M-11)		S			1000-1500
	M-18 (import from China)		S			700-1000
	Al Abid	Dev. Cancelled	L/L	48,000	2000	2000
Iraq	Tammuz 1	Dev. Cancelled			750	2000
	Badr 2000	Dev. Cancelled (Joint Iraq, Egypt/Argentina effort)			450	900
	Jericho 2B (YA-2B)		S/S			1300
Israel	Jericho 3	1990 IOC	S/S	29,000	1000	4800
	NoDong 1 (aka Rodong-1)	Flight Tested	L	21,000	800	1300
	Taepo Dong 1 (aka No Dong 2, Rodong-2)	In Dev. / Poss. 1 Fit Test	L/L	21,000	1000	1500-2000
North Korea	Taepo Dong 2	Engaged Effort	L/L		1000	3,500-6000
	Ghauri/Mk III	One flight test	L		500-750	1500
	CSS-2 (imported from China)	1988 IOC	L	65,200	1900	3100

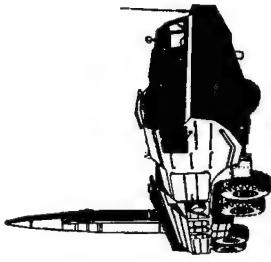
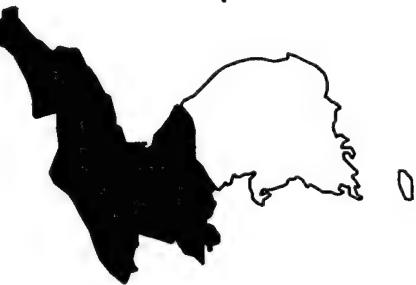
# Iran's Ballistic Missile Development Program



# Potential Iranian LRB<sup>M</sup> Coverage

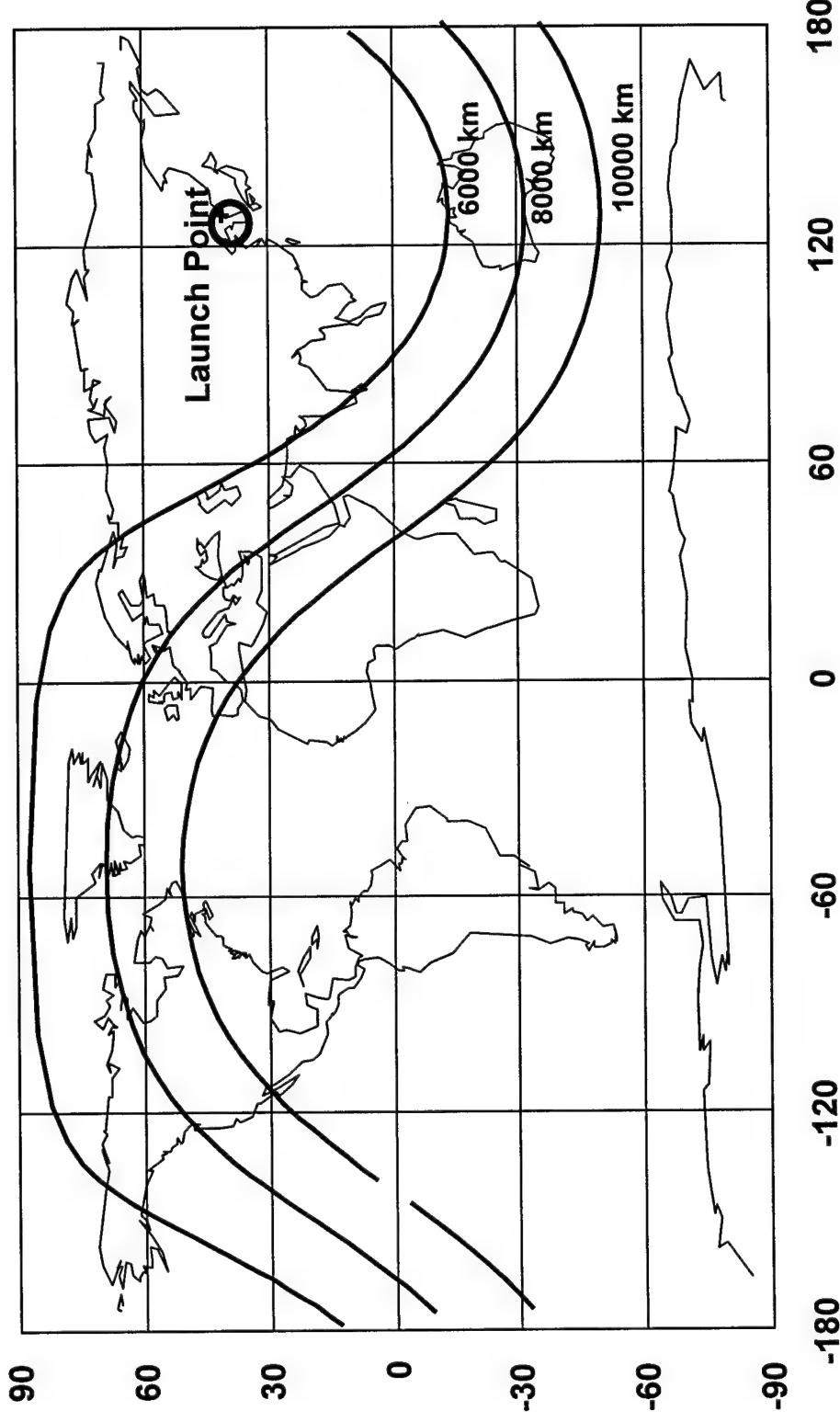


# North Korean Ballistic Missile Programs



- 1975 ~ DF-61 Program with China Terminated
- 1981 ~ Egypt Supplies Scud-B Missiles for Evaluation
- 1984 ~ Scud-B Mod A Tested
- 1985 ~ Scud-B Mod B Tested
- 1987 ~ Scud-B Mod B Exported to Iran
- 1989 ~ Scud-B Mod C Deployed
- 1990 ~ Nodong-1 Program Underway
- 1993 ~ Nodong-1 Tested
- 1995 ~ Taepo Dong 1 and 2 Programs Underway
- 1996 ~ Nodong-1 Exported to Iran
- 1998 ~ Nodong Technology in Pakistani Ghauri
- 1998 ~ Potential Space Launch/LRBM Capability

# Potential North Korean LRB<sup>M</sup> Coverage



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## **Outline**

---

- German WW II Missile Development and Technology Migration
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# **Overcoming Challenges to Third World LRBM Development**

---

## ***Obvious limitations...***

- **Access to Critical Technologies -- MTCR and economic sanctions**
  - **Engineering and manufacturing infrastructure -- old facilities**
  - **Missile system integration skills -- limited technical manpower**
- ... *mitigated by innovative shortcuts:*



# Key Technical Challenges

## Propulsion Systems

- Liquids
  - Ground handling and loading of propellants -- toxicity & volatility
  - Valves and flow control for predictable burnout
- Solids
  - Propellant grain consistency (motor-to-motor)
  - Dangerous large grain mix, pour, and cure
- Structures
  - Design, fabrication, and assembly of truss structures and interstages
  - Cluster/stack integration
  - Dynamic load margins throughout flight
- Control Electronics
  - Accurate navigation, timing, sensing of booster variations
  - Staging
- Power supplies and thermal control for LRBM duration flight

## Key Technical Challenges (Cont)

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- Reentry Systems
  - • Payload packaging for high deceleration shock loads
  - Stable dynamic shape design and balancing techniques
  - Thermal protection against heating loads at LRBM reentry velocities
- • Fusing for payload detonation/dispersal at desired altitude
- System Test and Verification Technology
  - Onboard system performance instrumentation
  - Telemetry electronics and antennas (ballistic missile and range)
- Flight data analysis tools

## 13 Critical Technologies

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- **High Energy Propellants**
- **Light Weight Subsystems**
- **Dynamic Structures Design and Analysis**
- **Aerodynamic Heat Protection Materials**
- **Advanced Flight Dynamics Control Systems**
- **Multi-Stage Separation and Ignition**
- **Payload Separation and Stabilization**
- **Payload Reentry Survival**
- **Warhead Fusing**
- **Guidance and Control Computers**
- **GPS/INS Platforms**
- **Flight Test Tracking and Telemetry**
- **High Fidelity Computer Simulation**

# Required Engineering Infrastructure

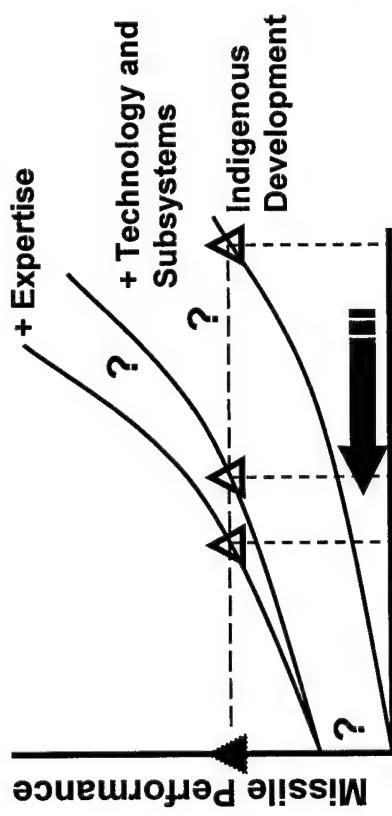
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- **Technical Expertise**
  - System design engineering
  - Fabrication methods
  - Materials & processes
  - Flight mechanics
  - Guidance & control
  - Modeling & simulation
- **Industry**
  - Propellant processing and handling
  - Propulsion subsystems
  - Electro/mechanical control systems
  - Structures manufacturing/assembly
  - Materials processing
  - Electronics production
  - Precision guidance fab. & assembly
  - Computer hardware and software
- **Flight Test**
  - Range size and access
  - Range instrumentation
  - Data analysis

**Home-Grown or “Contracted” Foreign Assistance Satisfies  
Required Development Functions**

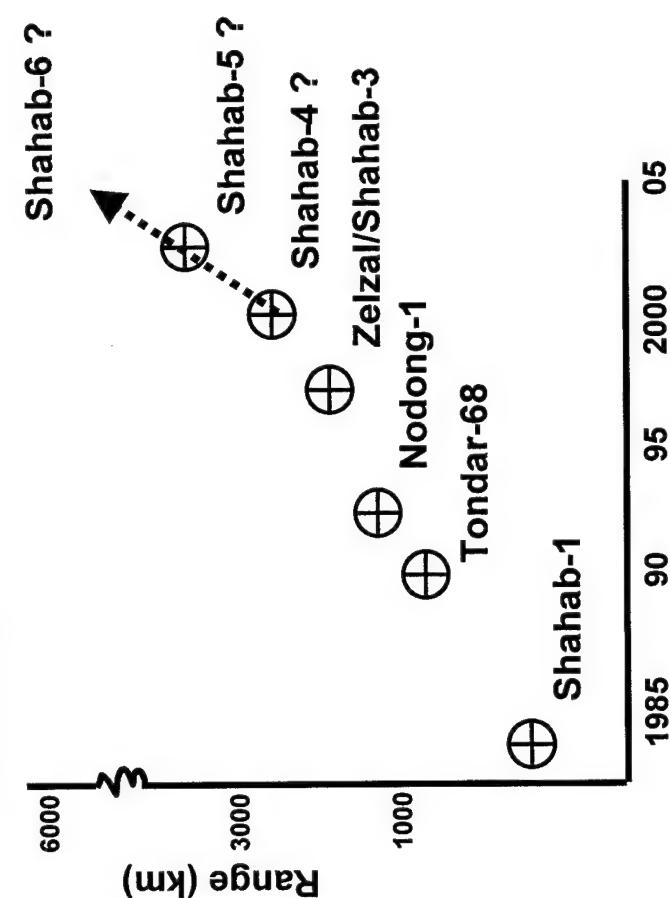
# Third World Development Time Compression

## Accelerating Missile Development



## Iran's Missile Program Evolution

- Combines imported systems with indigenous development and testing
- Extensive use of foreign expertise
- Steady range growth beyond regional requirements

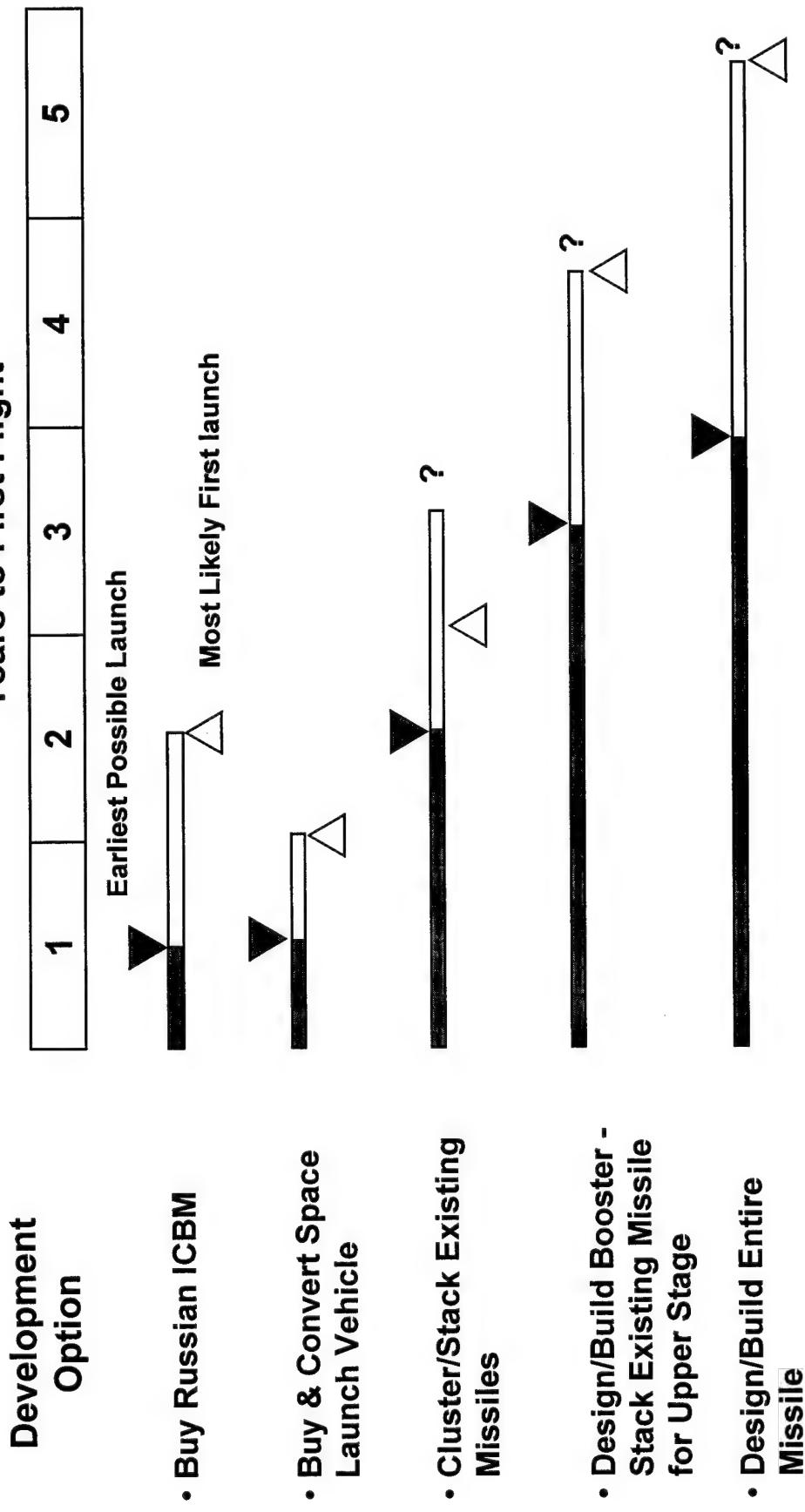


Desired Missile Capability  $\blacktriangle$  can be attained by a combination of development "investments" that may be hard to identify

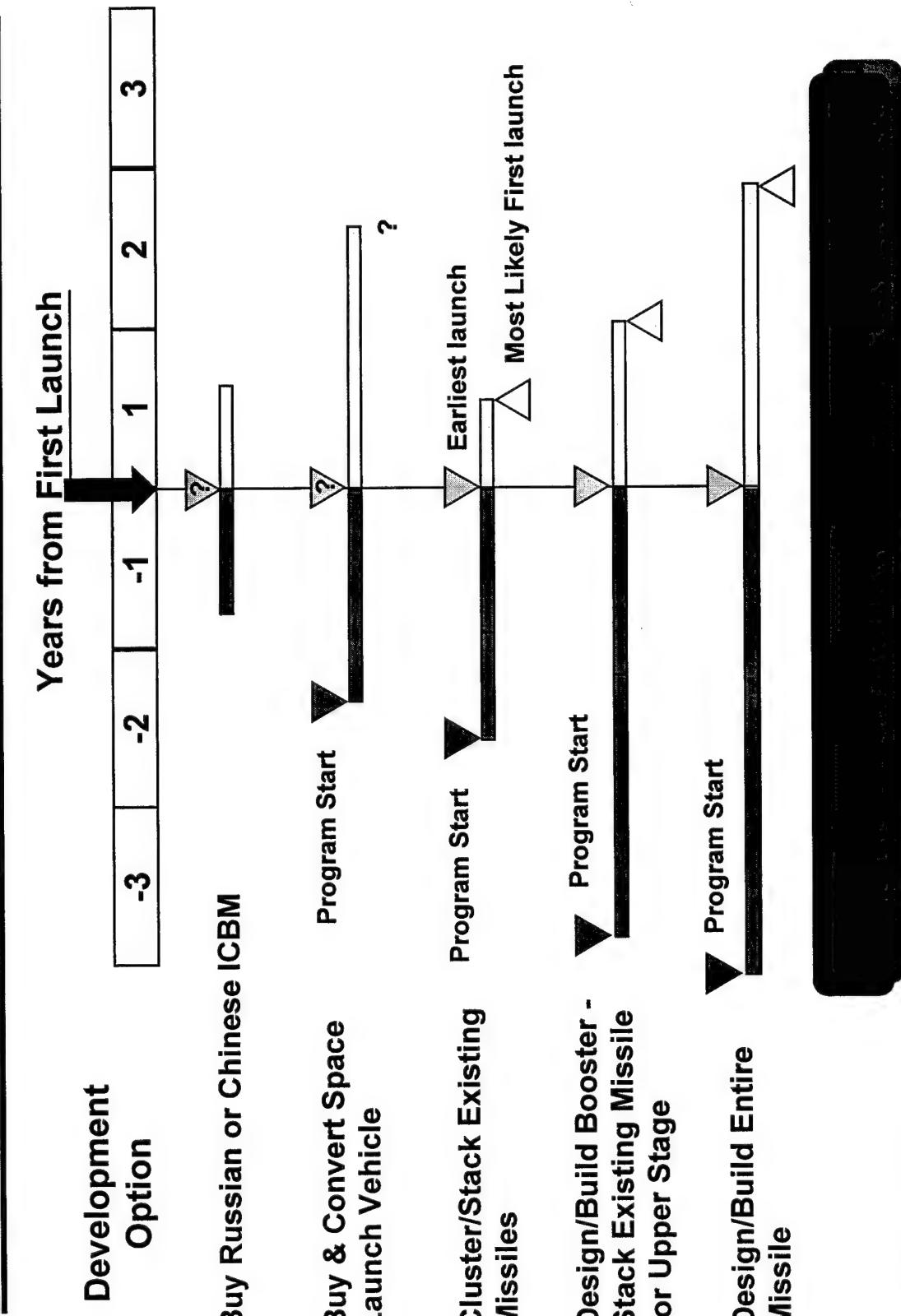
? Critical Unknowns:

- 1) Level of missile technology at "start point"
- 2) Foreign assistance to expedite system integration of acquired components

# Development Time Estimate



## **Effective Time to Respond**



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## **Options for LRB<sup>M</sup> Development**

---

- • **Buy LRB<sup>M</sup> (MRBM, IRBM, ICBM)**
- **Buy & Convert Space Launch Vehicle**
- **Cluster/Stack Existing Tactical Missiles**
- **Design/Build Booster - Stack Existing Missile for Upper Stage**
- **Design/Build Entire Missile**

## Available IRBMs

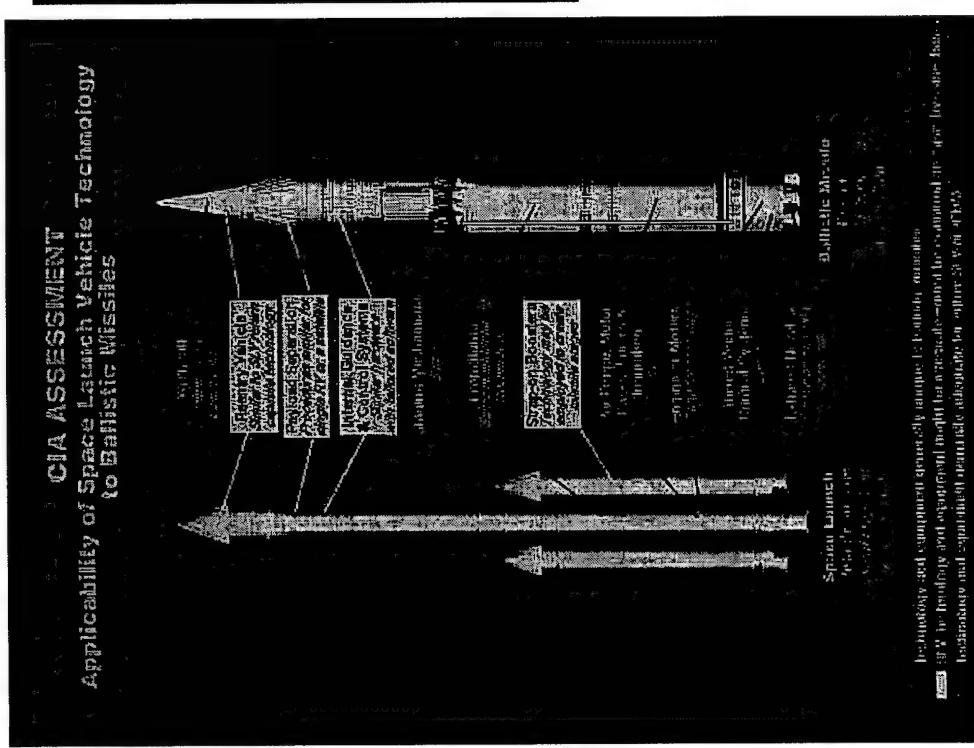
Missile	Payload(kg)	Range(Km)	Warhead	Country	IOC
• DF-3/DF-3A	2000/ 2150	2800	Conv./ NBC	China	1971/ 1986
• CSS-2	1900	3100	Conv.	Saudi Arabia	1988
• CSS-3	2200	4750	Conv./ NBC	China	1978
• DF-25	2000	1700	Conv./ NBC	China	Dev. Stop 1996
• Agni 1	1000	2000	Conv.	India	1998
• Agni 2	1000	3000	Conv./ NBC	India	2002?
• Shahab 4	1000	3000	Conv./ NBC	Iran	2002
• Shahab 5	?	?	Conv./ NBC	Iran	2005?
• Jericho 3	1000	4800	Conv./N?	Israel	2000?
• TD-1	1000	1500-2000	NBC	N.Korea	1998?
• TD-2	1000	2000-4000	NBC	N.Korea	1998?

## Options for LRB<sup>M</sup> Development

---

- Buy LRB<sup>M</sup> (MRBM, IRBM, ICBM)
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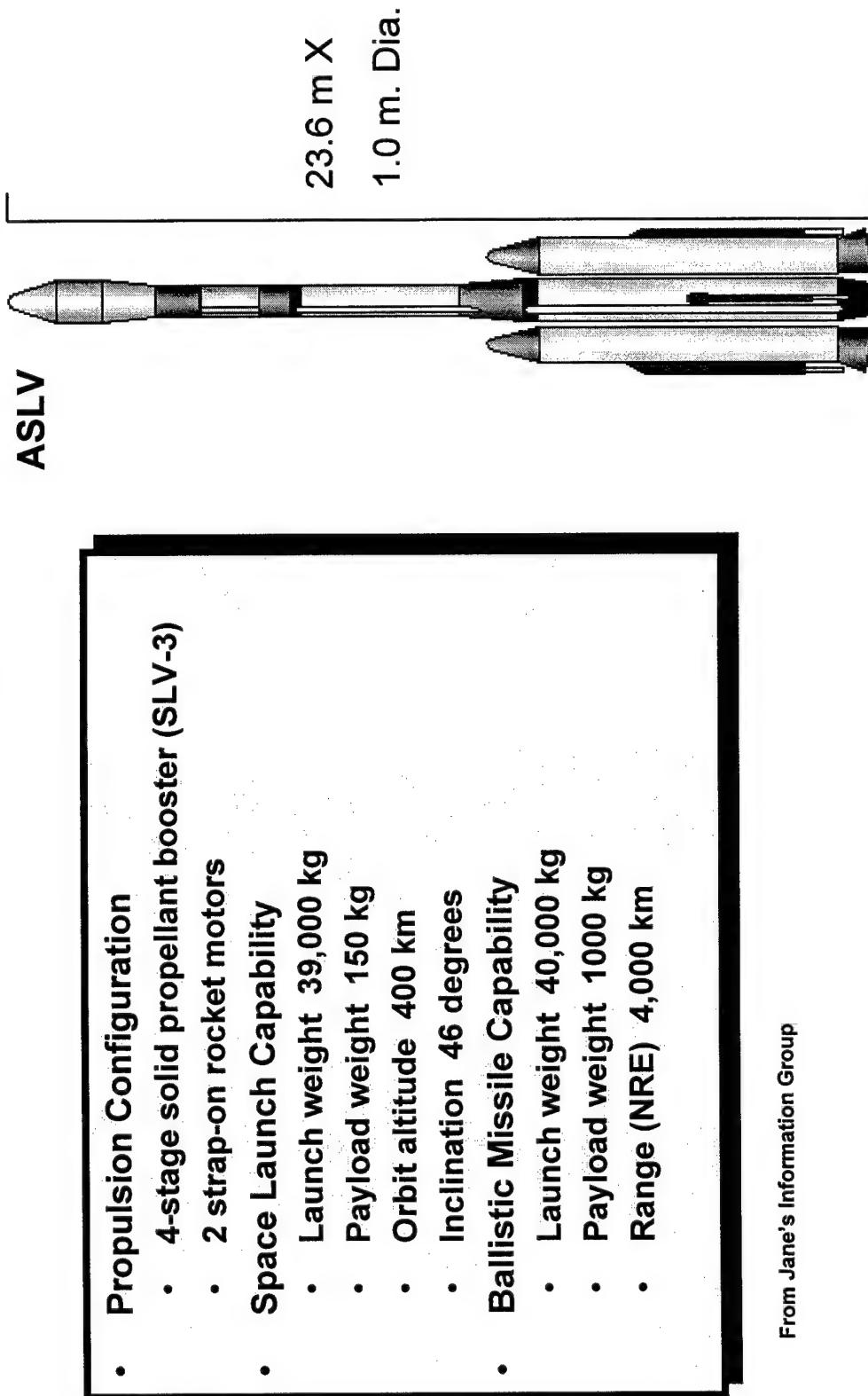
# CIA Assessment of SLV-LRBM Conversion



- Only unique ballistic missile technology is the warhead
- RV, separation, guidance & control, and strap-on booster SLV technologies may be adequate
- Staging, propellants, airframe, engines, thrust control, and nozzles are the same as SLV

Figure from 5/14/98 CIA briefing; as depicted in *Aviation Week & Space Technology*, June 1, 1998.

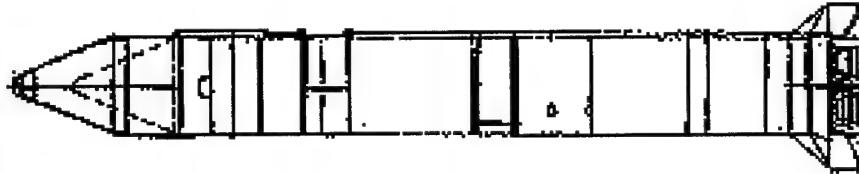
# Indian ASLV Conversion



From Jane's Information Group

# Israeli SHAVIT Conversion

SHAVIT SLV



- Propulsion Configuration
  - 3-stage solid propellant booster  
(based on Jericho 2)
  - NEXT, slightly larger follow-on
- Space Launch Capability
  - Launch weight 29,000 kg
  - Payload weight 800 kg
  - Orbit altitude 400 km
  - Inclination Polar
- Ballistic Missile Capability
  - Launch weight 29,000 (est.)
  - Payload weight 1,100 kg
  - Range (NRE) 5,000-7,000 km

14 m. X  
1.56 m. dia.

- Demonstrated 160 kg payload to 207 x 1587 km elliptical orbit @ 143 degree retrograde inclination.

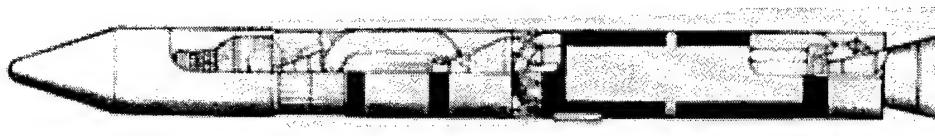
Jerico 2 From Federation of American Scientists web site:  
<http://www.fas.org/spp/guide/israel/launch/index.htm>

From Jane's Information Group

# Japanese M-3 or M-5 Conversion

- The M-3 space launch vehicle “family” could be converted into an IRBM with a 500+ kg payload and a range of 4,000+ km. The Japanese Government has officially refuted this allegation. \*

## M-5 SLV



- The M-5 is a three- or four-stage, solid propellant launch vehicle designed to carry payloads of 2,000 kg to 200 Km; 1,200 kg to 500 km and 800 kg to GTO. \*
- Three-stage version shown:
  - Length = 31.0 m
  - Diameter = 2.5 m
  - Launch weight = 130,000 kg

*M-5 LRBM performance has not yet been assessed*

From Jane's Information Group

MORSS June 1999

Source: Jane's Strategic Weapons

## **Options for LRB<sup>M</sup> Development**

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- Buy LRB<sup>M</sup> (MRBM, IRBM, ICBM)
- Buy & Convert Space Launch Vehicle
- Cluster/Stack Existing Missiles
  - Design/Build Booster - Stack Existing Missile for Upper Stage
  - Design/Build Entire Missile

# Two Feasible LRBM Designs

## Liquid

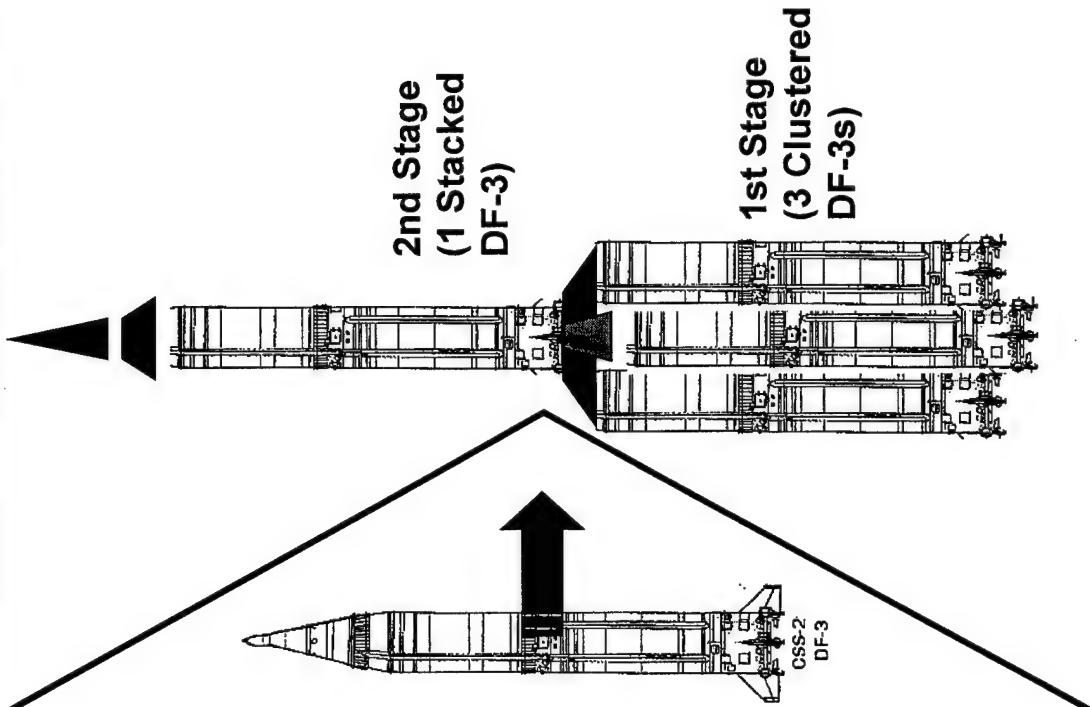
- Clustered CSS-2 ballistic missiles
    - Stage 1 -- 3 CSS-2's } Separate
    - Stage 2 -- 1 CSS-2 } at burnout
- Payload -- 1000 kg  
Guidance -- 150 kg  
Launch Weight -- 255,400 kg  
Reentry Angle -- 30 degrees  
Time of Flight -- 46.5 minutes  
Range -- 10,000 km

## Solid

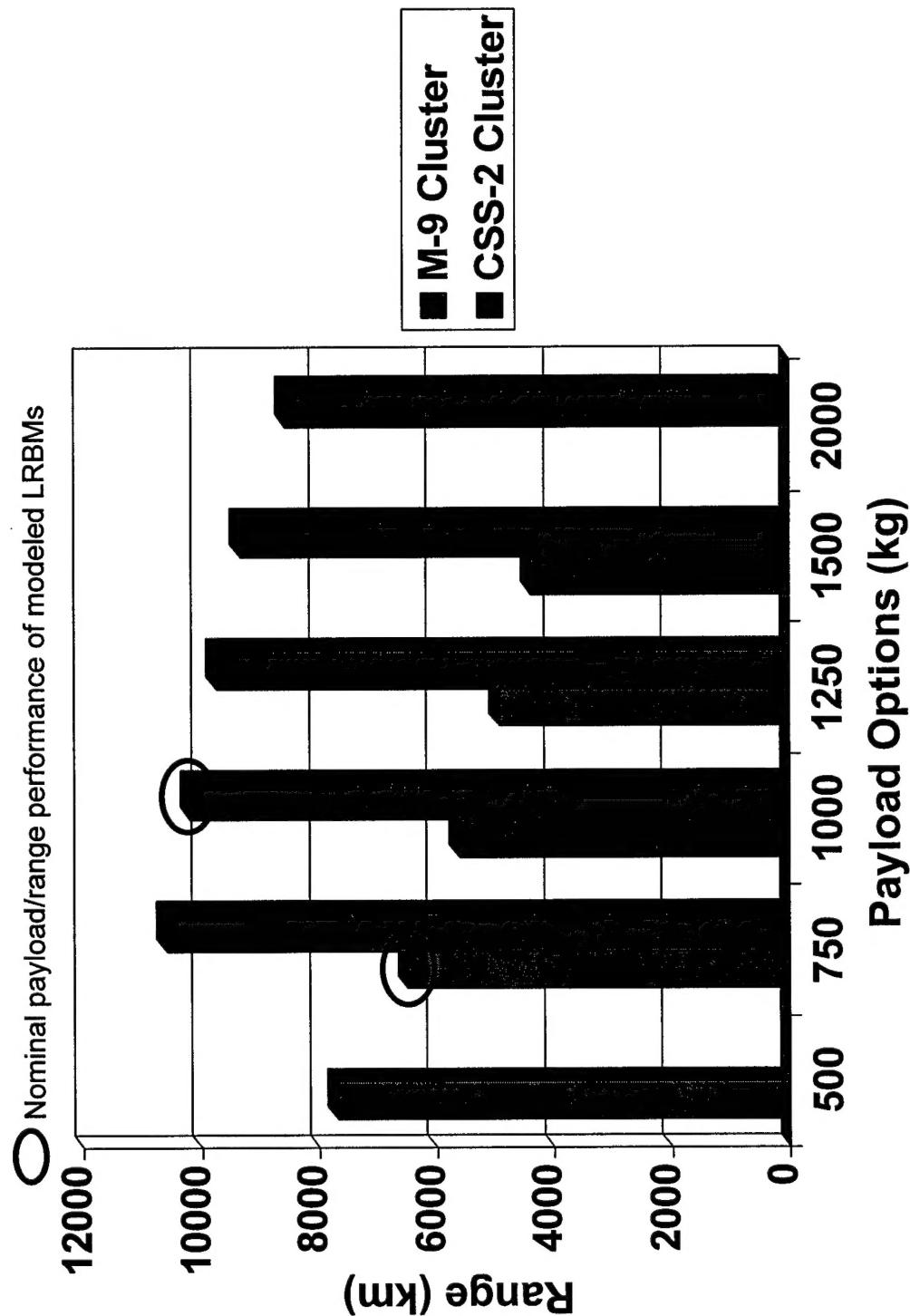
- Clustered M-9 ballistic missiles
    - Stage 1 -- 7 M-9's }
    - Stage 2 -- 3 M-9's }
    - Stage 3 -- 1 M-9 }
- Separate at burnout
- Payload -- 750 kg  
Guidance -- 150 kg  
Launch Weight -- 59,000 kg  
Reentry Angle -- 30 degrees  
Time of Flight -- 28.5 minutes  
Range -- 6,500 km

# Design Issues Considered in Assessment

- RV structure & materials for 30 degree reentry
- Payload separation at booster burnout; G&C stays with spent upper stage
- Interstages between booster stages sized for flight loads at increased diameter
- Revised raceways and electrical system wiring
- Guidance subsystem increased power and cooling
- Truss structure for clustered rocket motors
- Extensible nozzle exit cones (upper stages only)
- Booster separation mechanisms
- Booster thermal protection for aero loads and motor nozzle heat



# Range Sensitivity to Payload Weight



# Evaluation Methodology

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- **AS-2530 Flight Simulation Code**
  - 3 - degrees of freedom
  - Non-rotating spherical earth
  - Standard atmospheric properties
- **Powered ascent flight profile**
  - Vertical launch
  - Instantaneous pitchover
  - Gravity turn
  - Payload separation at booster burnout
- **Payload flight profile**
  - Keplerian free flight
  - Keplerian reentry
  - No reentry angle of attack
- **Missile characteristics**
  - Nominal ballistics
  - Nominal weights
- **Estimated drag properties**

# **Outline**

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- German WW II Missile Development and Technology Migration
- Trends in Third World Ballistic Missile Weaponry
- Threat Development on a Compressed Schedule
- Candidate LRBM Configurations
- Summary

## **Summary**

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- Third World Countries are capable of achieving LRBM capability
- Innovative technical shortcuts may be used for different national goals
  - Political or coercive vs military capability
- Time to First Flight may be much earlier than currently anticipated
- Time from observed threat to first launch may be very short
- MTCR is not stopping missile technology proliferation
- Developed countries are currently marketing critical ballistic missile technologies, systems, and personnel

**The Time Between Inescapable Evidence of a Developing Threat to When That Threat is Fielded May be Less Than the Time in Which We Can Respond**